

ASEN 5052-001, 5052-001B Analytical Astrodynamics

TTh, 10:05-11:20, AERO 111

Instructor: Daniel Scheeres, scheeres@colorado.edu
AERO 454
Office Hours TBD

Introduction to astrodynamics with an emphasis on analytical approaches — alternative to ASEN 5050. General solution of the 2-body problem. Orbital trajectories, transfers, targeting, and time of flight. Orbit perturbations and averaging analysis. Restricted 3-body problem.

Pre-requisite: Undergraduate orbital mechanics course (equivalent to ASEN 3200) or permission of the instructor.

Coursepack:

Selected excerpts from “Orbital Motion in Strongly Perturbed Environments” will be distributed, selected papers will be distributed.

Textbooks:

A.E. Roy, Orbital Motion 4th edition, Institute of Physics Publishing, 2005.

Additional Reference Books:

D.J. Scheeres. “Orbital Motion in Strongly Perturbed Environments: Applications to Asteroid, Comet and Planetary Satellite Orbiters,” Springer-Praxis Books in Astronautical Engineering. 2012. ISBN 978-3-642-03255-4, e-ISBN 978-3-642-03256-1, DOI 10.1007/978-3-642-03256-1

Grading:

HW problems:	25%
Computational problems:	25%
Mid-term exam:	25%
Final exam:	25%

Topics:

Principles of orbital mechanics.
Orbital trajectories, transfers, time of flight.
Trajectory propagation and targeting.
Orbit perturbation formulation and analysis.
Restricted 3-body problem with applications.

Syllabus (Scheeres):

Orbital mechanics

Formulation of two-body, three-body and n-body problems
The two-body problem solution
Elliptical and circular orbits
Parabolic and hyperbolic trajectories
3-D trajectories and orbit elements
Time of flight and orbit propagation

Orbital transfers

Impulsive maneuvers
Lambert's theorem
3-D Targeting
Fuel optimal considerations

Orbit perturbation formulations

Variation of constants
Lagrange's Equations
Gauss' Equations
Mean elements and averaging

Orbit perturbation analysis

Effect of non-spherical gravity fields
Low-thrust trajectories
Atmospheric drag
Tidal and third body effects

Restricted 3-body problem with applications

Derivation of equations of motion
Jacobi Integral, Zero-Velocity Curves, and Lagrange Points
Hill approximation
Numerical computation and analysis of orbits!

In-Class vs Remote course access:

The following items detail my plans for delivering lectures and office hours, accommodating any restrictions that may arise from the current pandemic crisis. If the campus transitions to a more restrictive stage, the course has been designed to be able

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: ; G0A4; E: ; 5L''% =4AE<H4; '4; 'A: X8: 7H; F'<>>4EE49<H4; 7'07'4><5: 9'4; '56: '&C7<D0705@'(: AGC>: 7'K: D705: L'
04; 5<>5'&C7<D0705@'(: AGC>: 7'<5']_]TbV^TcadU'4A'97C; =4e>4?4A<94L: 98'=4A'=8A56: A'<77C75<; >; L''%@48'6<G: '<
5: EB4A<A@'E: 9C<?>'4; 9C0H4; J'7: : '-: EB4A<A@' : 9C<?>'04; 9C0H4; 7'4; '56: '&C7<D0705@'(: AGC>: 7'K: D705: L'

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<; 9'BA4; 48; 7'<A: '7C75: 9'4; 'C; 75A8>54A7g'><77'A475: A7L'% '56: '<D7: ; >; '4='78>6'8B9<5: 7J'56: '; <E: '56<5'
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'C; >?89: J'D85'<A: '; 45'7CE5: 9'54i'B?<FC<AC7EJ'>6: <H; FJ'=<DAC><H4; J'?@C; FJ'DACD: A@'56A: <5J'
8; <8564ACf: 9'<>>: 77'54'<<<9: EC>'E<5: AC<?7J'>'C>Y: A'=A<89J'78DECI ; F'56: '7<E: '4A'7CEC?<A'K4AY'C; '
E4A: '56<; '4; : '>48A7: 'K056485'B: AEC77C4; '=A4E'<??'>48A7: 'C; 75A8>54A7'C; G4?G: 9J'<; 9'<C9C; F'
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j64; 4Ae>4?4A<94L: 98kh']_]TbV^Tl l l _kl'(589: ; 57'=-48; 9'A: 7B4; 7CD?: '=4A'GC4?<H; F'56: '<<<9: EC>'
'C; 5: FAC5@'B4?C>@'K0??'D: '78DM: >5'54'; 4; <<<9: EC>'7<; >H4; 7'=A4E'56: '24; 4A'049: '<7'K: ??<7'
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EC7>4; 98>5'j6<A<77E: ; 5J': 0B'4C5<H4; J'<; 9'<77<8'5KJ'C; HE<5: 'B<A5; : A'GC4?: ; >; 'j9<H; F'4A'
94E: 7H>'GC4?: ; >; kJ'75<'YC; FJ'4A'BA45: >5: 9T>?<77'9C7>ACEC; <H4; '4A'6<A<77E: ; 5'D@'4A'<F<C; 75'
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<; 9'=<A?'K56'<?'7589: ; 57'K64J'D: ><87: '4='A: ?FC487'4D?CF<H